

# Today's Topics

Friday, February 7, 2025 (Week 2, lecture 7) – Chapter 3.

0. Gravity review

1. Circular Motion

*... Newton's version of Kepler's 3rd law.*

2. Center of Mass

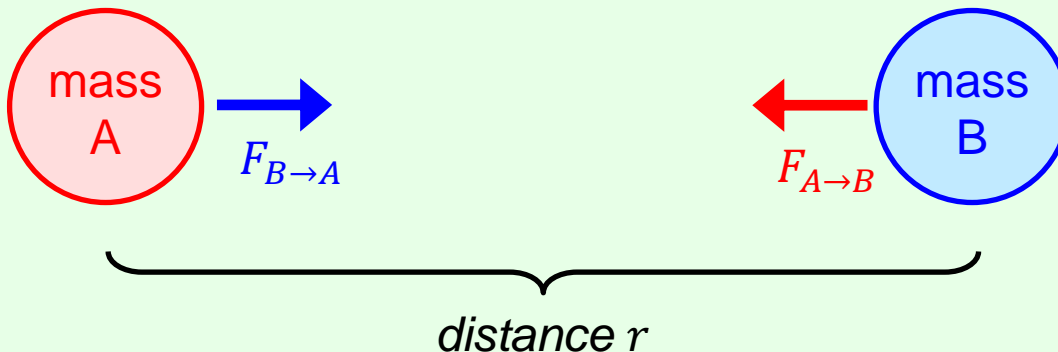
3. Angular momentum

# Gravity Review

## Newton's law of universal gravitation

All masses attract each other according to the following relation:

$$F_{A \rightarrow B} = -G \frac{M_A M_B}{r^2} = -F_{B \rightarrow A}$$



### Properties

- Falls off as  $1/r^2$ .
- Proportional to  $M_A$ .
- Proportional to  $M_B$ .
- $G =$  Newton's constant  
 $= 6.67430(15) \times 10^{-11}$   
 $m^3 / Kg \cdot s^2$

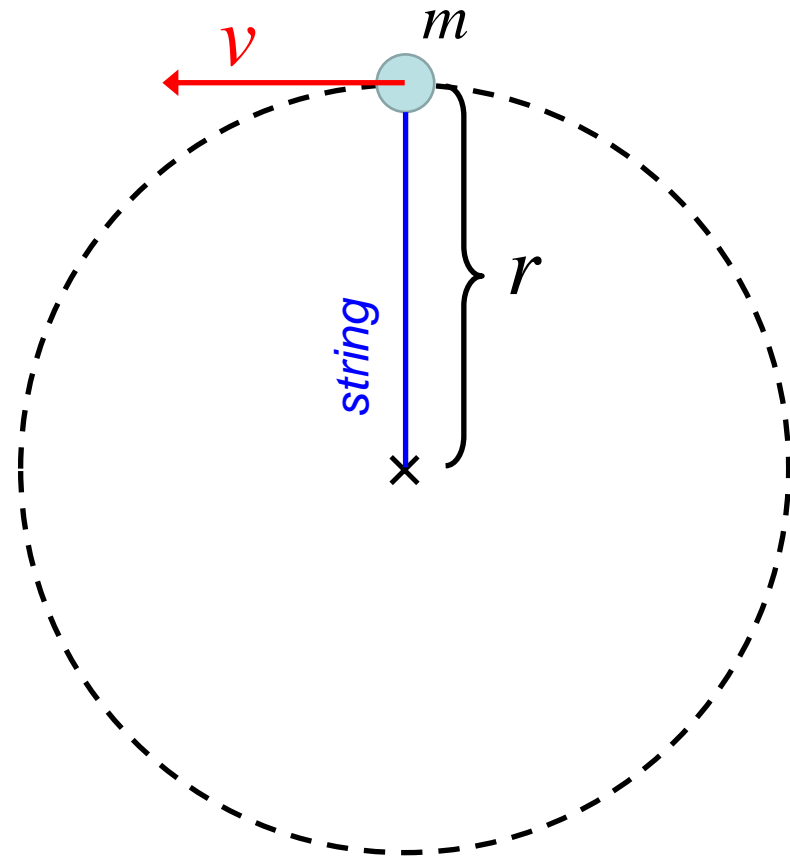
**Why do all objects** (on Earth)  
**fall**  
**at the same rate?**

# Circular Motion

## Recall

acceleration = **change** in **velocity** over time  
*speed & direction*

**Rotation** is a type of acceleration where the velocity **direction changes**, but speed is constant.



“ball on a string”

# Circular Motion

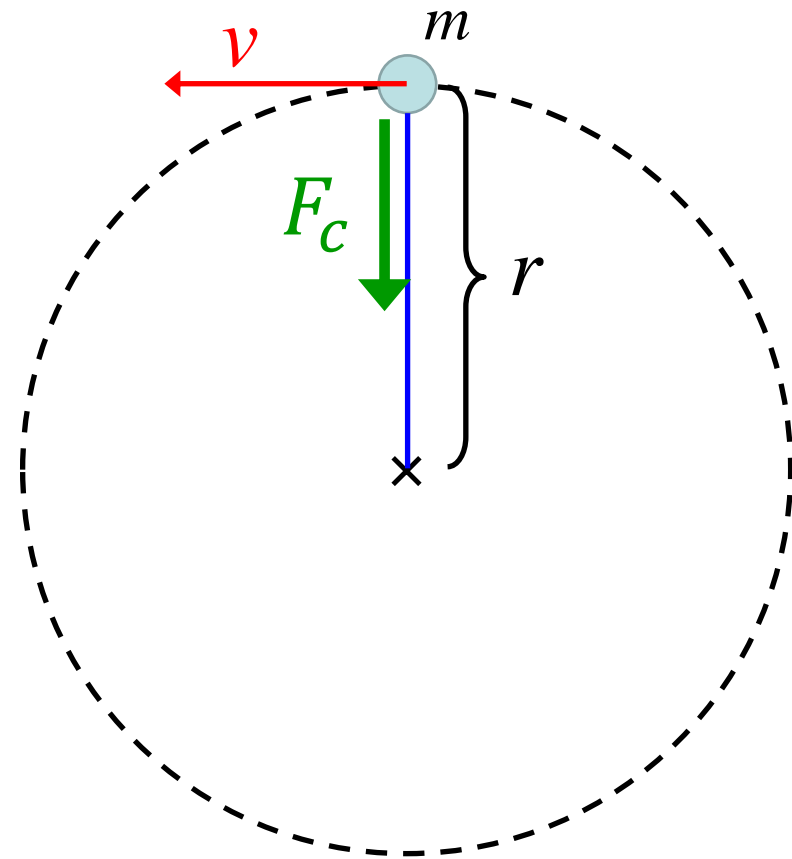
## Recall

acceleration = **change** in velocity over time  
*speed & direction*

**Rotation** is a type of acceleration where the velocity **direction changes**, but speed is constant.

$$\text{Acceleration: } a_c = \frac{v^2}{r}$$

$$\text{Centripetal Force: } F_c = \frac{mv^2}{r}$$

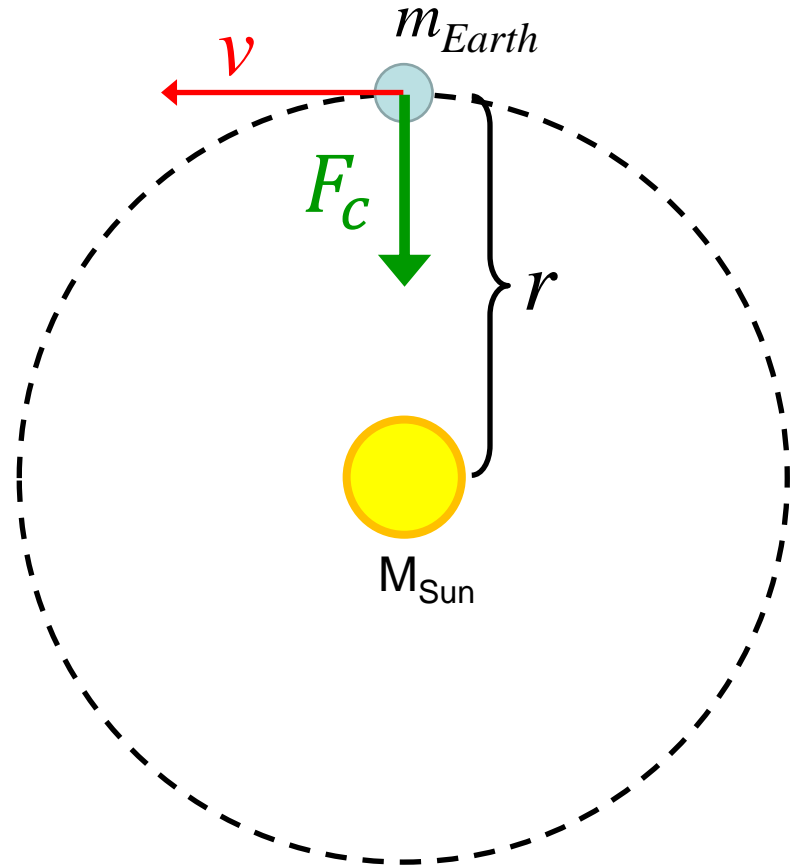


“ball on a string”

# Circular Motion Example: Earth's orbit of Sun

Centripetal force needed to keep  
Earth on a circular orbit:

$$F_c = \frac{m_{Earth} v_{Earth}^2}{r}$$



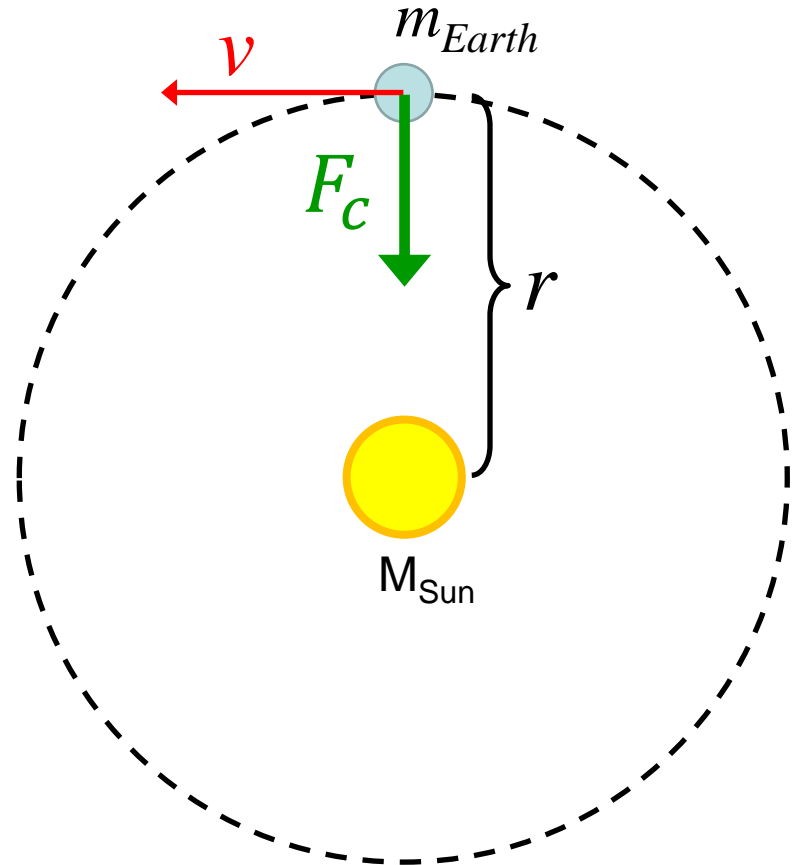
# Circular Motion Example: Earth's orbit of Sun

Centripetal force needed to keep  
Earth on a circular orbit:

$$F_c = \frac{m_{Earth} v_{Earth}^2}{r}$$

Force of gravity on Earth from Sun:

$$F_{gravity, S \rightarrow E} = G \frac{m_{Earth} M_{Sun}}{r^2}$$



# Circular Motion Example: Earth's orbit of Sun

Centripetal force needed to keep Earth on a circular orbit:

$$F_c = \frac{m_{Earth} v_{Earth}^2}{r}$$

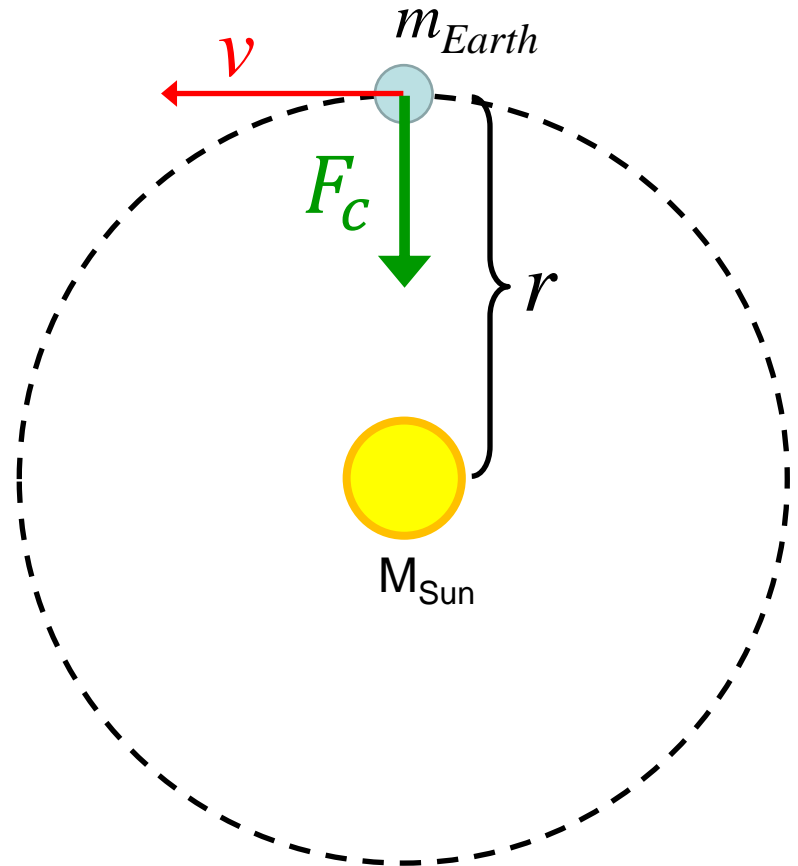
Force of gravity on Earth from Sun:

$$F_{gravity, S \rightarrow E} = G \frac{m_{Earth} M_{Sun}}{r^2}$$

The **centripetal force** that pulls on Earth to make it orbit the Sun **is gravity**:

$$F_c = F_{gravity, S \rightarrow E}$$

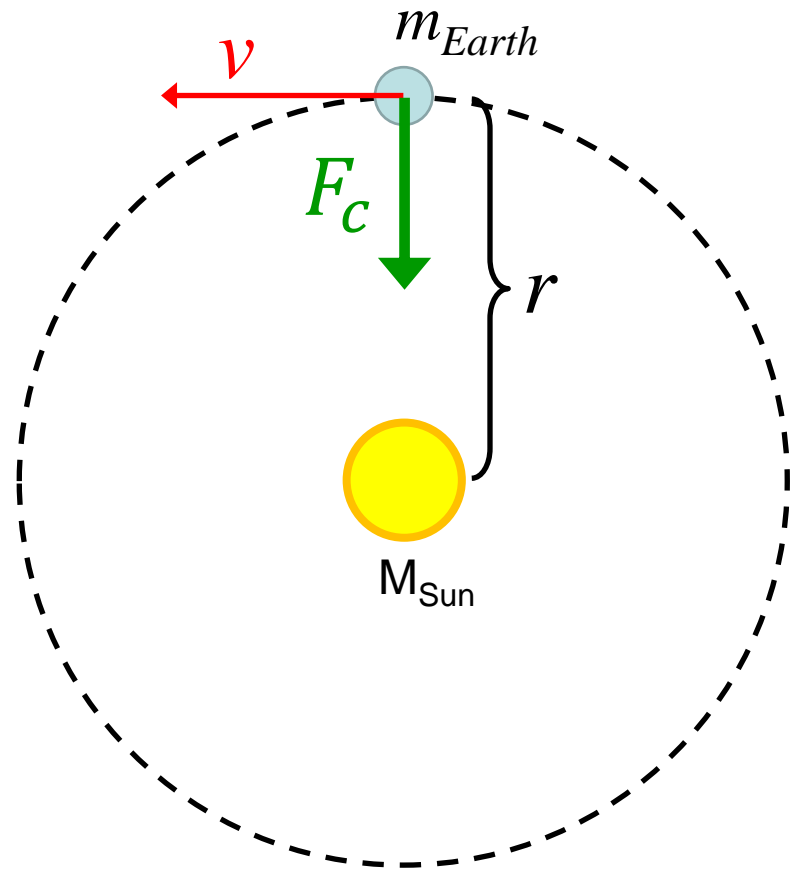
$$\Leftrightarrow \frac{m_{Earth} v_{Earth}^2}{r} = G \frac{m_{Earth} M_{Sun}}{r^2}$$





# Circular Motion Example: Earth's orbit of Sun

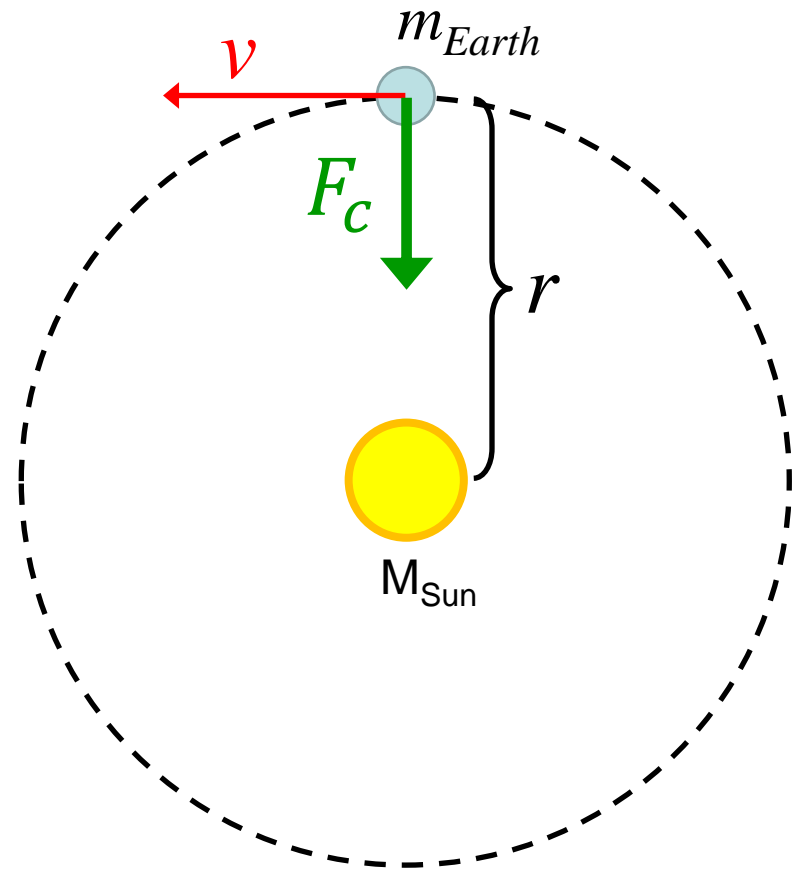
$$\frac{m_{\text{Earth}} v_{\text{Earth}}^2}{r} = G \frac{m_{\text{Earth}} M_{\text{Sun}}}{r^2}$$



# Circular Motion Example: Earth's orbit of Sun

$$\frac{\cancel{m_{\text{Earth}}} v_{\text{Earth}}^2}{r} = G \frac{\cancel{m_{\text{Earth}}} M_{\text{Sun}}}{r^2}$$

$$\Leftrightarrow v_{\text{Earth}}^2 = G \frac{M_{\text{Sun}}}{r}$$



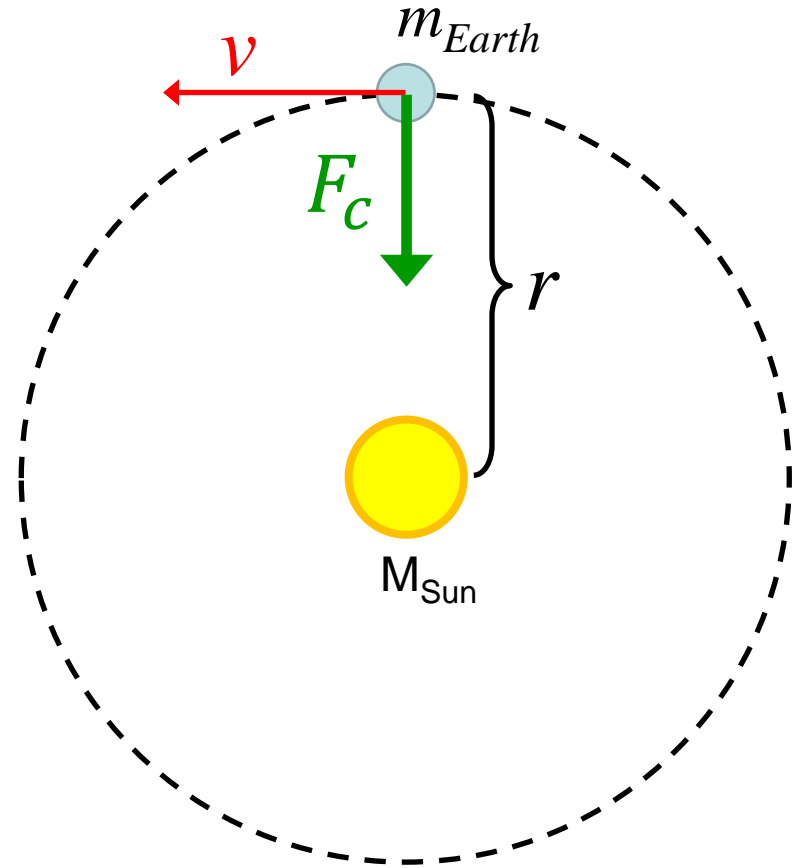
# Circular Motion Example: Earth's orbit of Sun

$$\frac{\cancel{m_{Earth}} v_{Earth}^2}{r} = G \frac{\cancel{m_{Earth}} M_{Sun}}{r^2}$$

$$\Leftrightarrow v_{Earth}^2 = G \frac{M_{Sun}}{r}$$

Solve for  $M_{Sun}$  :

$$M_{Sun} = \frac{r v_{Earth}^2}{G}$$



# Circular Motion Example: Earth's orbit of Sun

$$\frac{\cancel{m_{Earth}} v_{Earth}^2}{r} = G \frac{\cancel{m_{Earth}} M_{Sun}}{r^2}$$

$$\Leftrightarrow v_{Earth}^2 = G \frac{M_{Sun}}{r}$$

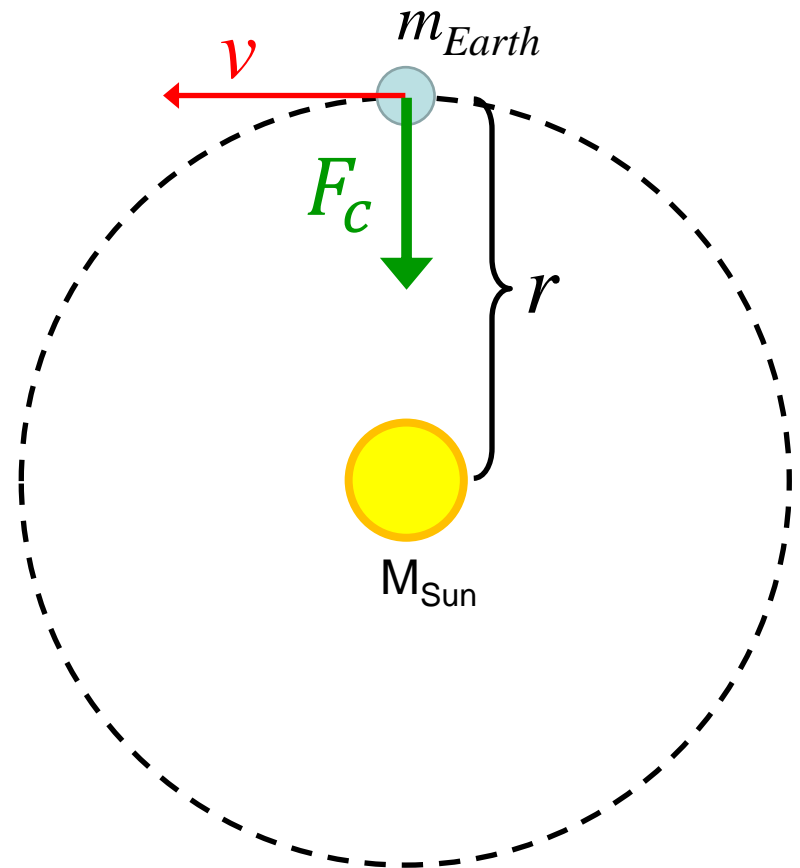
Solve for  $M_{Sun}$  :

$$M_{Sun} = \frac{r v_{Earth}^2}{G}$$

$$v_{Earth} = 29.78 \times 10^3 \text{ m/s}$$

$$r = 1 \text{ AU} = 149.6 \times 10^9 \text{ m}$$

$$G = 6.67430(15) \times 10^{-11} \text{ m}^3/\text{Kg} \cdot \text{s}^2$$



# Circular Motion Example: Earth's orbit of Sun

$$\frac{\cancel{m_{Earth}} v_{Earth}^2}{r} = G \frac{\cancel{m_{Earth}} M_{Sun}}{r^2}$$

$$\Leftrightarrow v_{Earth}^2 = G \frac{M_{Sun}}{r}$$

Solve for  $M_{Sun}$  :

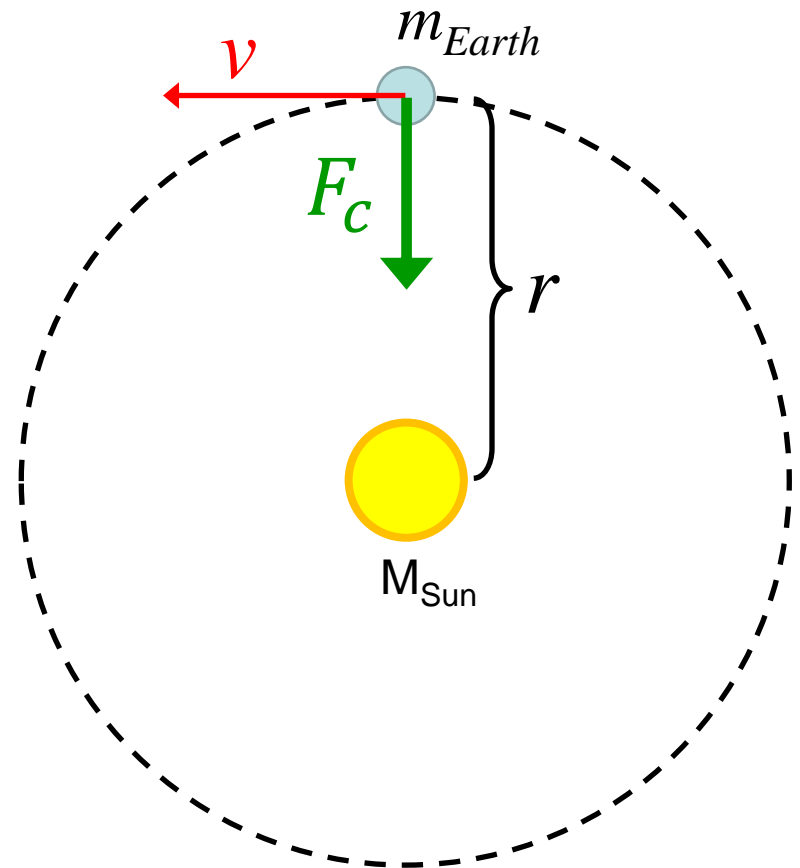
$$M_{Sun} = \frac{r v_{Earth}^2}{G}$$

$$v_{Earth} = 29.78 \times 10^3 \text{ m/s}$$

$$r = 1 \text{ AU} = 149.6 \times 10^9 \text{ m}$$

$$G = 6.67430(15) \times 10^{-11} \text{ m}^3/\text{Kg} \cdot \text{s}^2$$

$$M_{Sun} = 1.988 \times 10^{30} \text{ Kg}$$



You can get the mass of the Sun from  
Earth's orbital parameters !!!

# Newton's version of Kepler's 3rd Law

$$T^2 = \frac{4\pi^2}{G(M_1 + M_2)} a^3$$

*This formula is in SI units*

T = orbital period in seconds

$M_{1,2}$  = Mass of orbiting objects in Kg

a = semimajor axis in meters

G =  $6.6743 \times 10^{-11} \text{ m}^3/\text{Kg}\cdot\text{s}^2$

# Newton's version of Kepler's 3rd Law

$$T^2 = \frac{4\pi^2}{G(M_1 + M_2)} a^3$$

*This formula is in SI units*

T = orbital period in seconds

$M_{1,2}$  = Mass of orbiting objects in Kg

a = semimajor axis in meters

G =  $6.6743 \times 10^{-11} \text{ m}^3/\text{Kg}\cdot\text{s}^2$

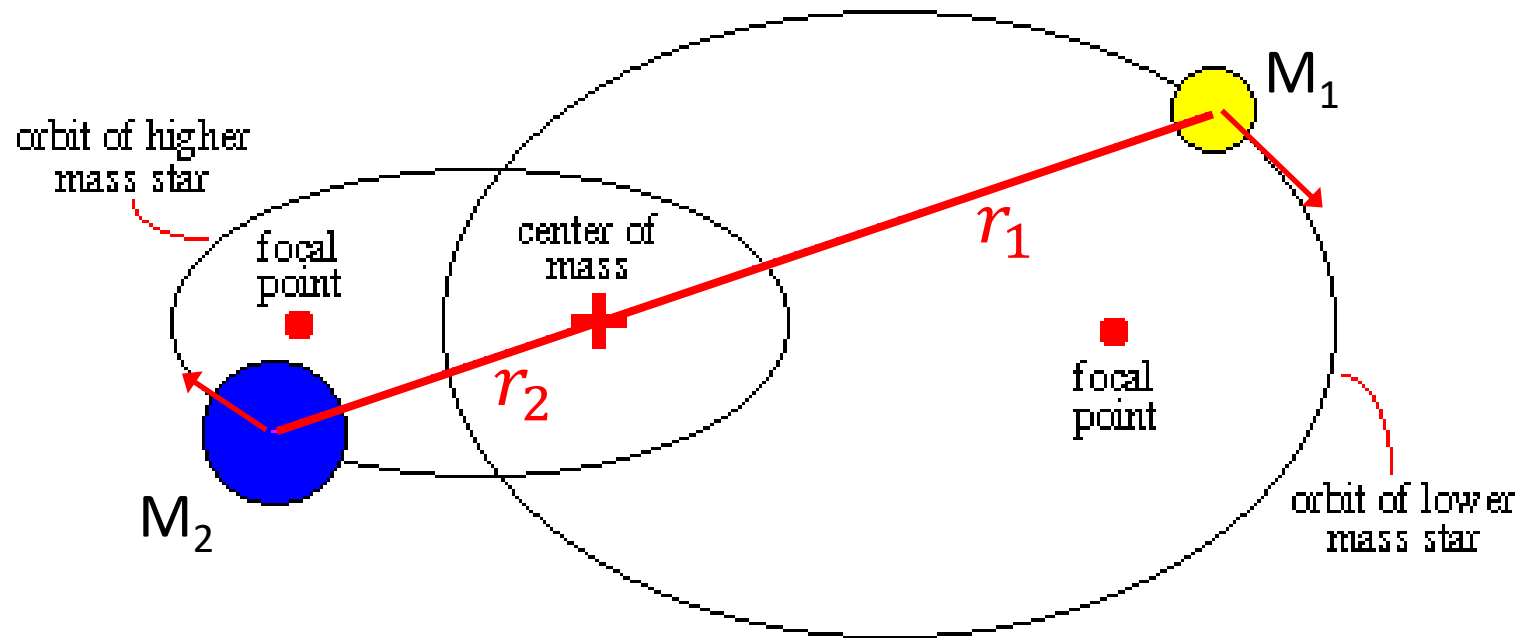
**WHAT IF:** What happens to the orbits if  $M_1$  and  $M_2$  are comparable ?

# Center of Mass



# What happens when $M_1 \simeq M_2$ ?

The **center of mass** of  $M_1$  and  $M_2$  serves as the orbiting ellipse focus.

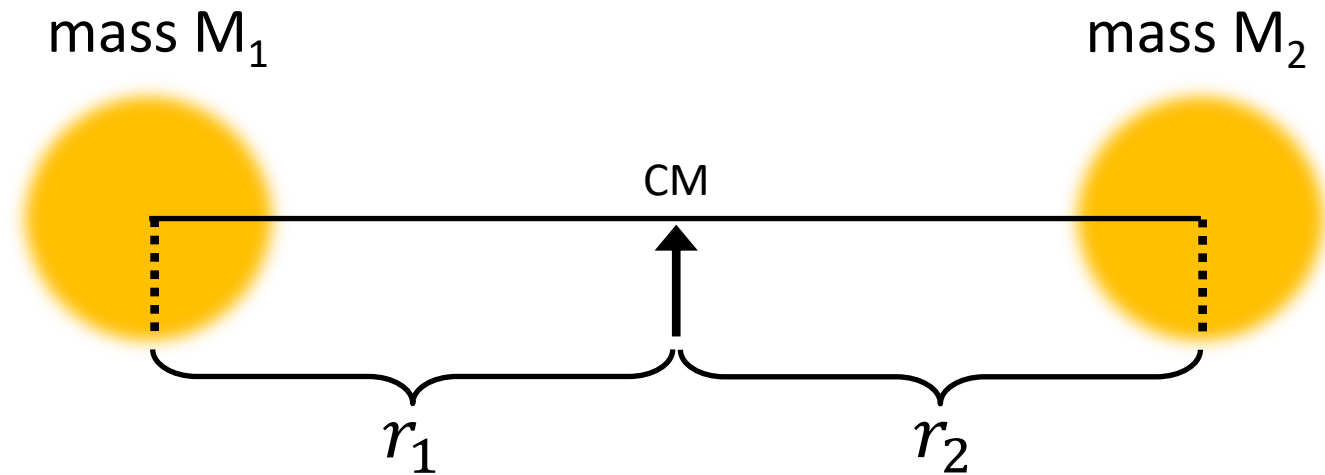


[adapted from <http://abyss.uoregon.edu>]

## Semimajor axis “a”:

The coordinate “ $r = r_1 + r_2$ ” is the distance between the two masses. It also describes an ellipse (not shown), whose semimajor axis “a” is used in Newton’s version of Kepler’s 3rd law.

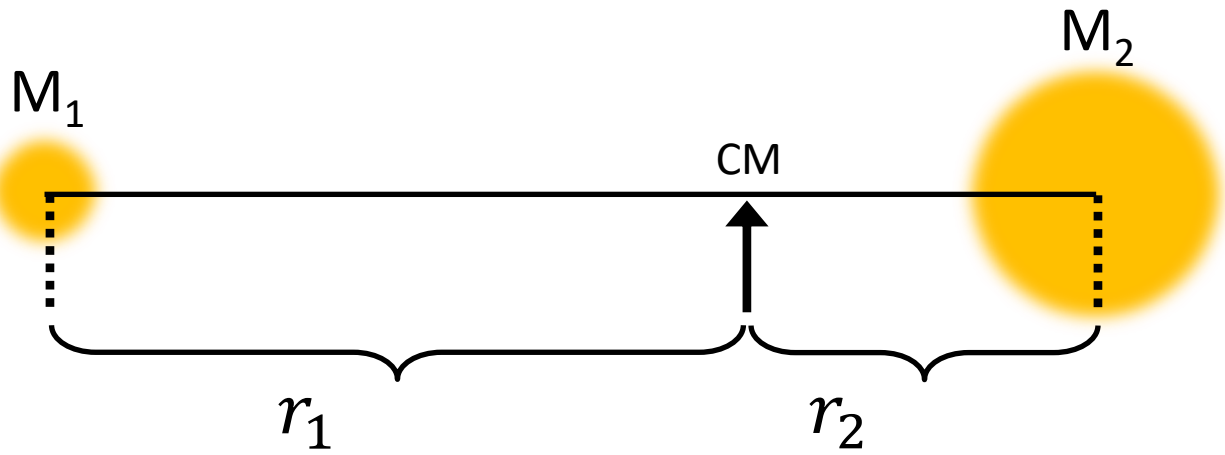
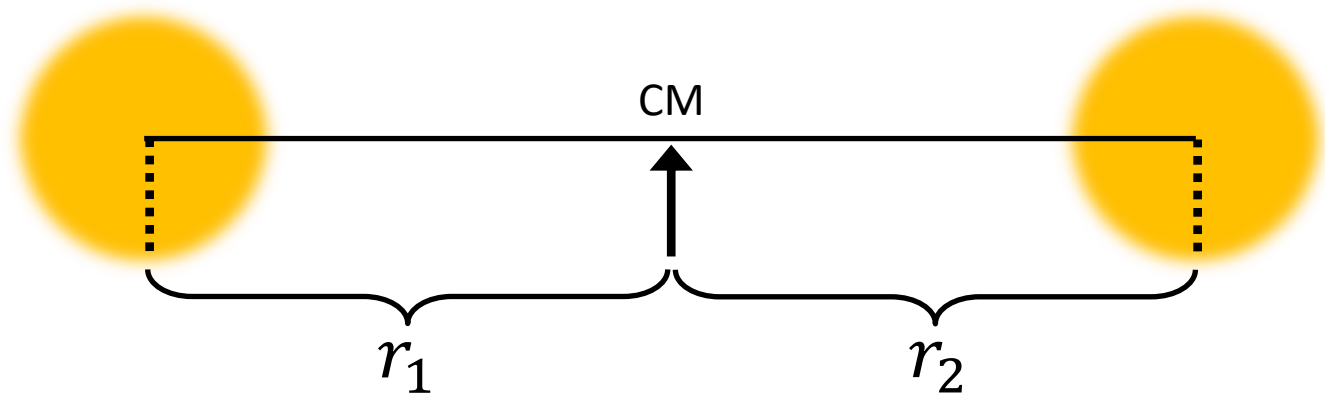
# Center of Mass



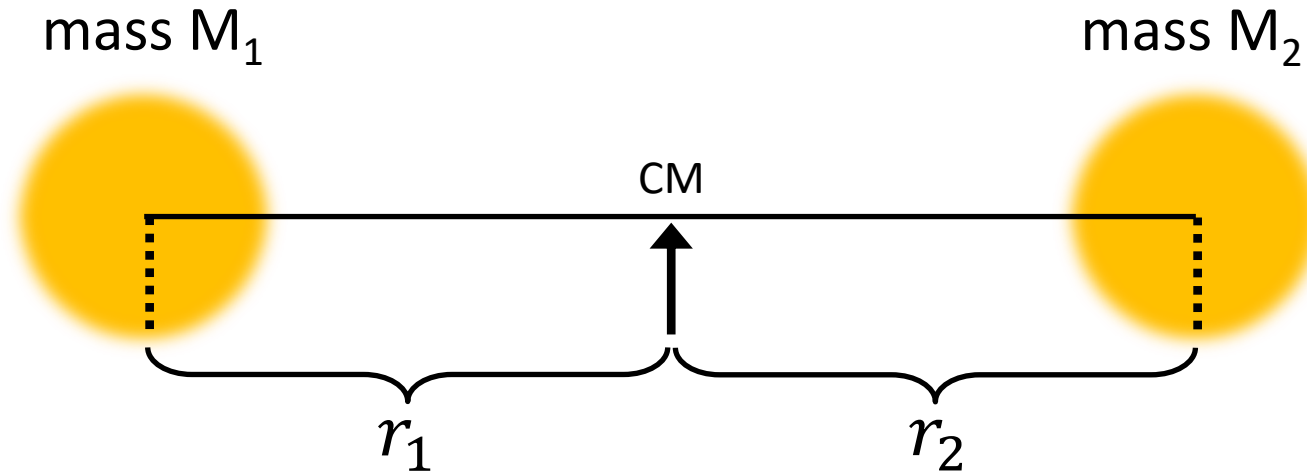
# Center of Mass

mass  $M_1$

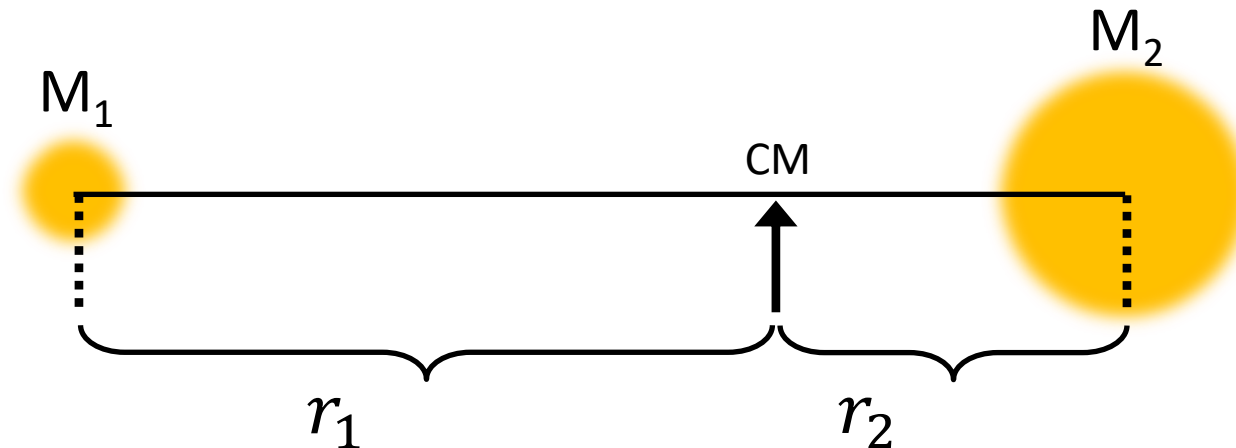
mass  $M_2$



# Center of Mass



Center of mass is located such that  $M_1 r_1 = M_2 r_2$   
(or "barycenter")



# Some Barycenters

$M_2 - M_1:$   $r_2 = a \frac{M_1}{M_1 + M_2} = \text{distance from CM to } M_2$

**Sun-Earth:**  $r_2 = 448 \text{ km} = 3.0 \times 10^{-6} \text{ AU}$

**Earth-Moon:**  $r_2 = 4,670 \text{ km}$  with  $a = 384,000 \text{ km}$   
 $= 73\% \text{ of Earth's radius}$

# Some Barycenters

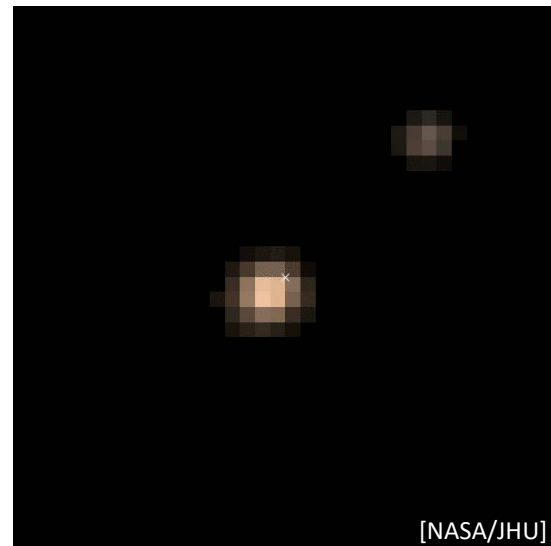
$M_2 - M_1:$   $r_2 = a \frac{M_1}{M_1 + M_2} = \text{distance from CM to } M_2$

**Sun-Earth:**  $r_2 = 448 \text{ km} = 3.0 \times 10^{-6} \text{ AU}$

**Earth-Moon:**  $r_2 = 4,670 \text{ km}$  with  $a = 384,000 \text{ km}$   
 $= 73\% \text{ of Earth's radius}$

**Pluto – Charon:**

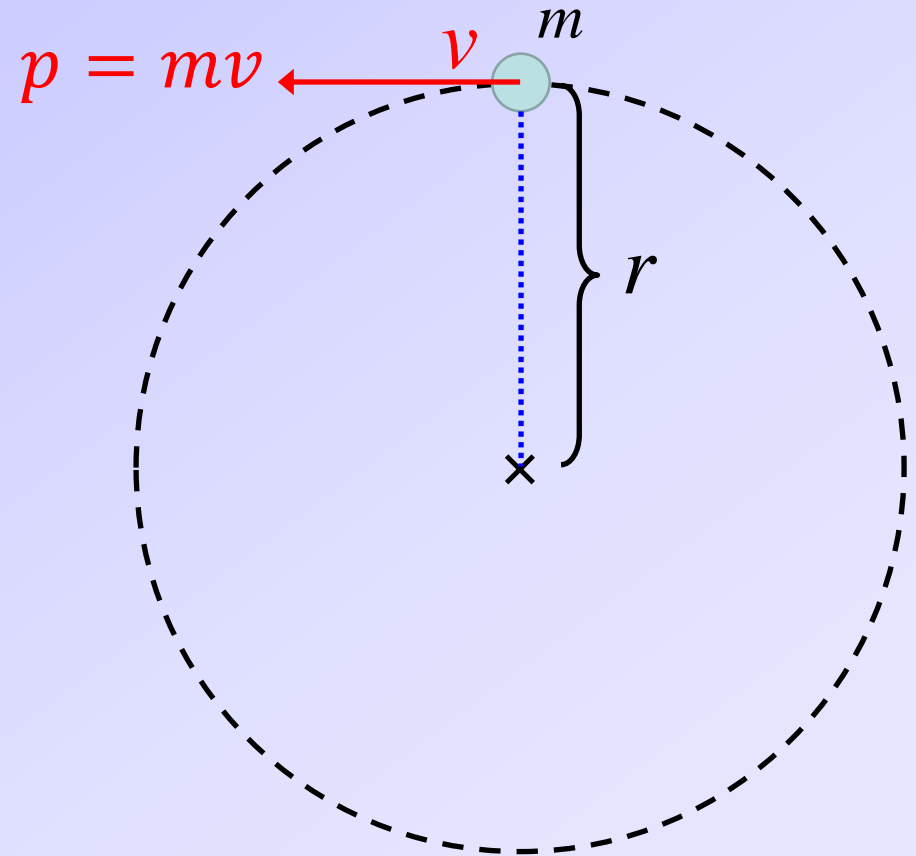
*orbital period  $T = 6.4 \text{ days}$*



**PolleEv Quiz: [PolleEv.com/sethaubin](https://PolleEv.com/sethaubin)**

# Conservation of Angular Momentum (1)

angular momentum =  $L = \text{momentum} \times \text{radius}$   
 $= p \times r \quad \dots = mvr$  for circular motion

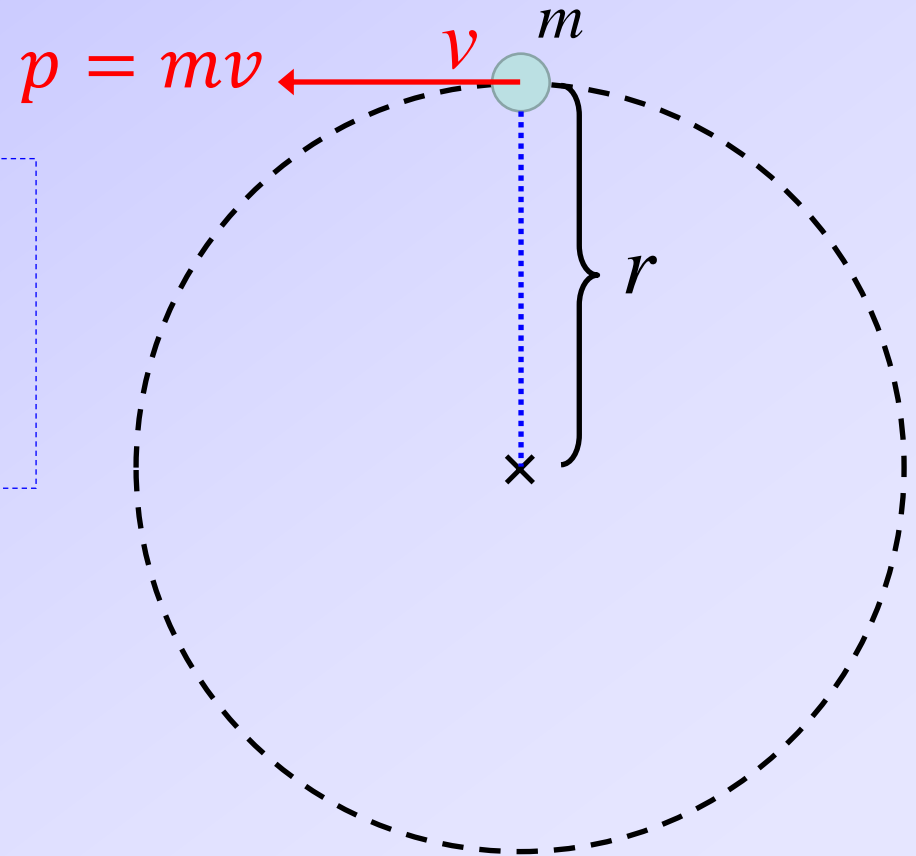




# Conservation of Angular Momentum (1)

angular momentum =  $L = \text{momentum} \times \text{radius}$   
 $= p \times r \quad \dots = mvr$  for circular motion

***total angular momentum***  
=  
sum of the angular momenta of  
all the sub-parts of a system



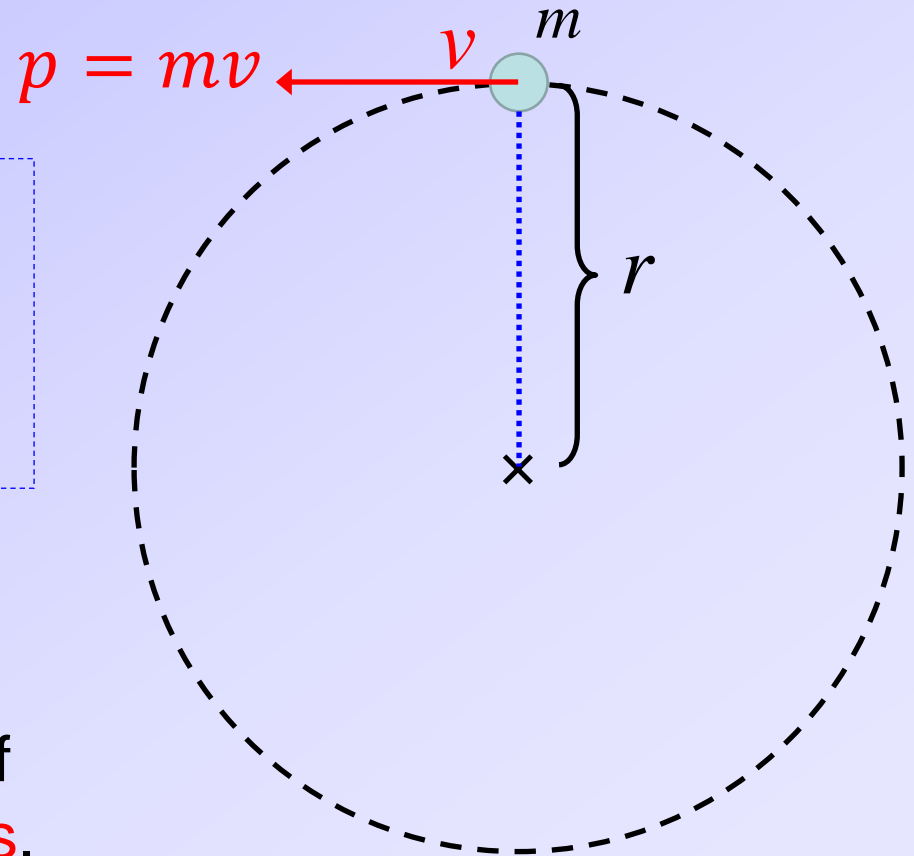
# Conservation of Angular Momentum (1)

angular momentum =  $L = \text{momentum} \times \text{radius}$   
 $= p \times r \quad \dots = mvr$  for circular motion

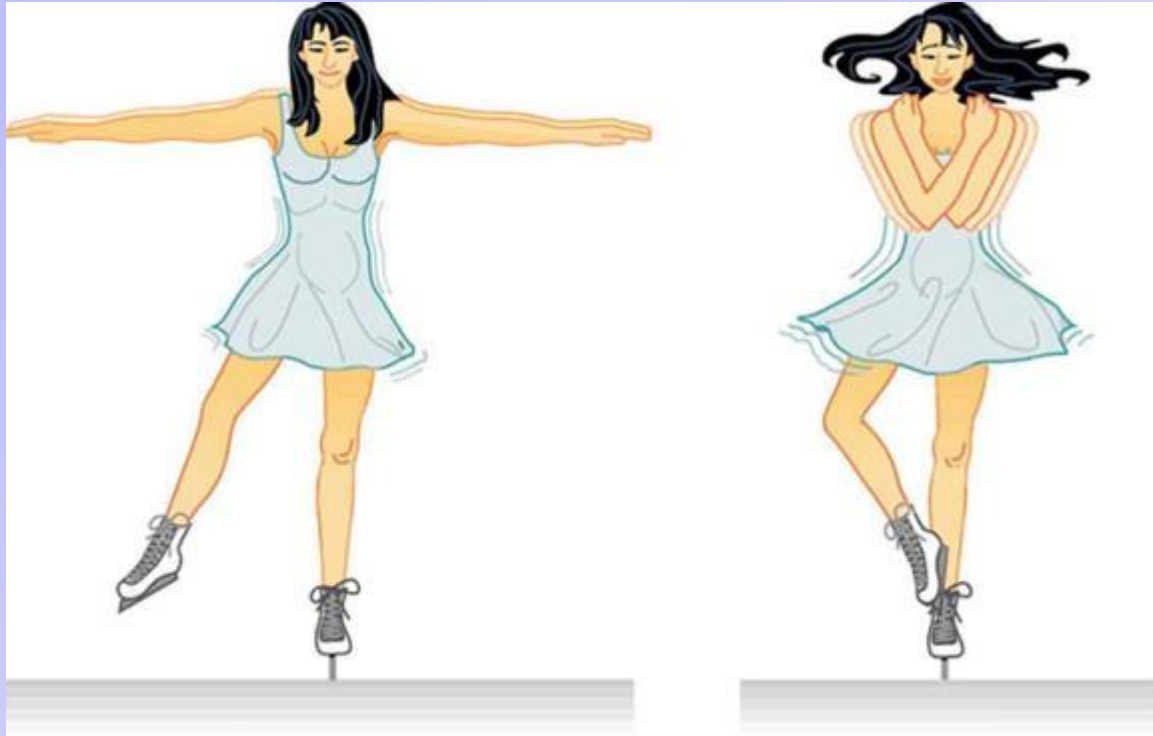
***total angular momentum***  
=  
sum of the angular momenta of  
all the sub-parts of a system

## Conservation Law

The **total angular momentum** of  
a **closed system** **never changes**.



# Conservation of Angular Momentum (2)



[OpenStax: Astronomy]

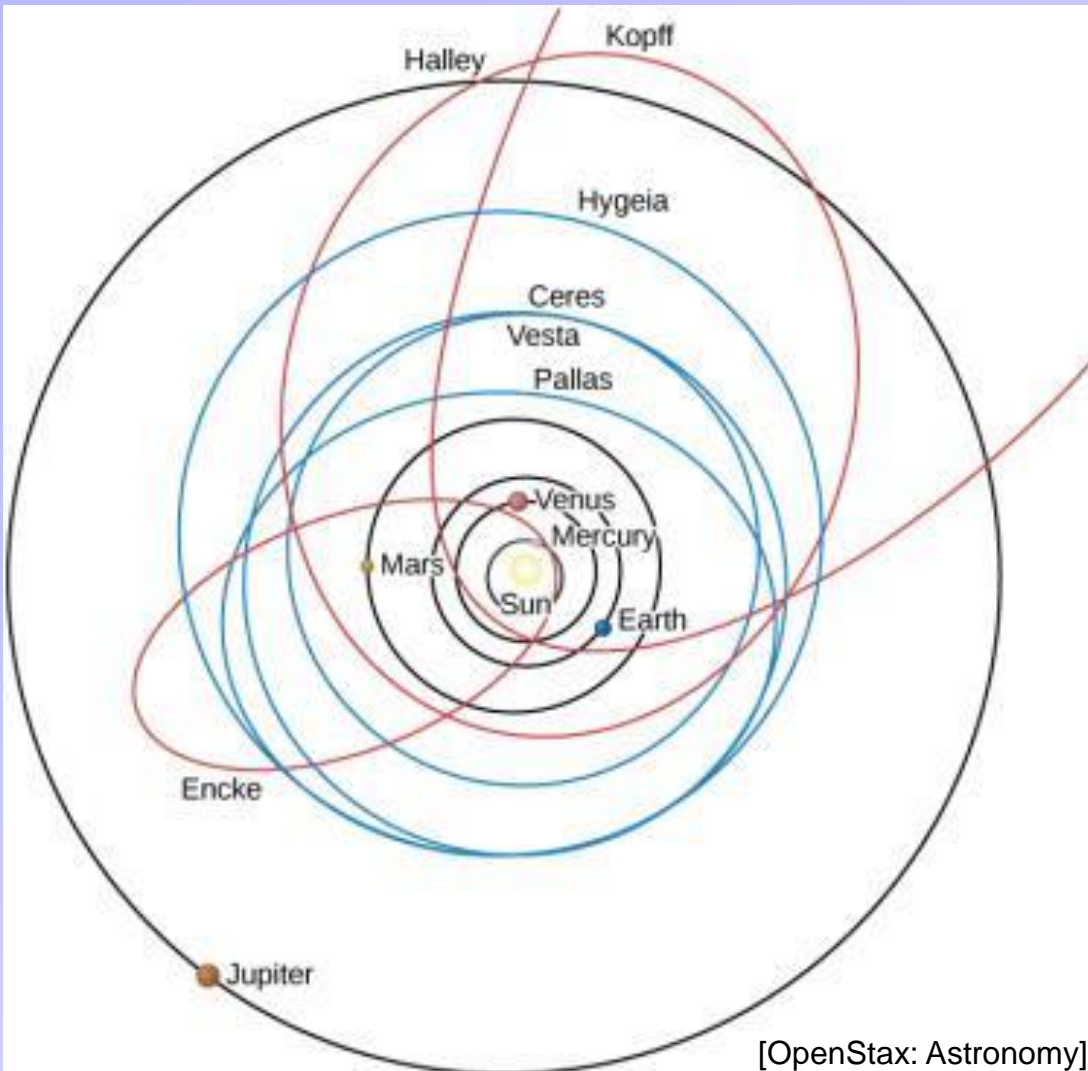
- When a spinning figure skater **brings in her arms**, their distance from her spin center is **smaller**, so her **speed increases**.
- When her **arms are out**, their distance from the spin center is **greater**, so she **slows down**.

# Conservation of Angular Momentum

The multiple planets, asteroids, and comets all interact and modify each others orbits.

→ **Individual angular momenta change.**

→ **Total angular momentum of Solar System is constant.**



Planets (black), asteroids (blue), comets (red)

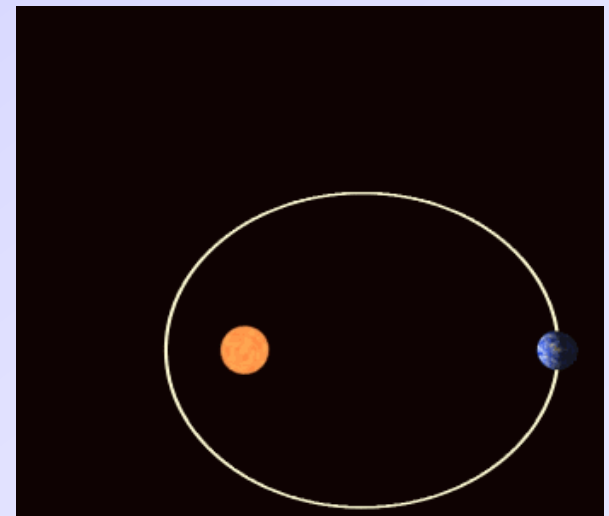
# Conservation of Angular Momentum

The multiple planets, asteroids, and comets all interact and modify each others orbits.

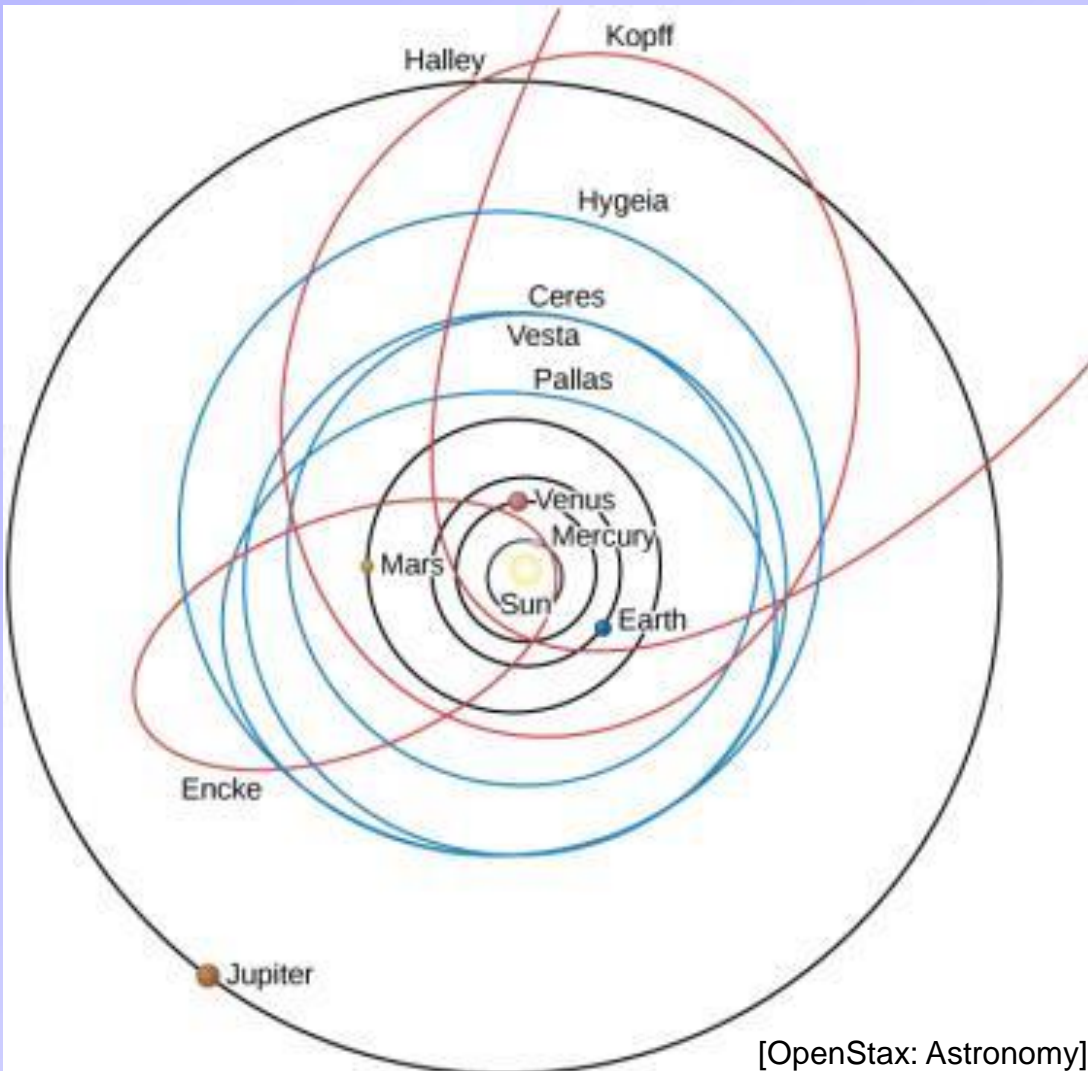
→ **Individual angular momenta change.**

→ **Total angular momentum of Solar System is constant.**

Example: Apsidal Precession



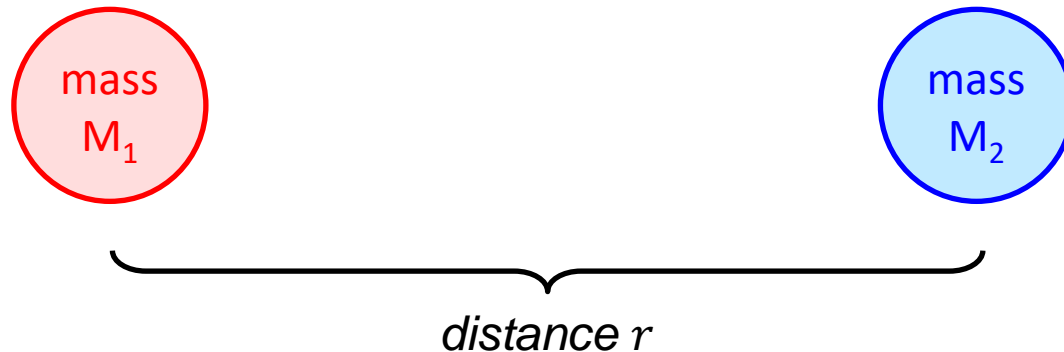
By WillowW - Own work, CC BY 3.0,  
<https://commons.wikimedia.org/w/index.php?curid=3416065>



[OpenStax: Astronomy]

Planets (black), asteroids (blue), comets (red)

# Gravitational Potential Energy



$$\text{Stored gravitational energy} = E_{potential} = -G \frac{M_1 M_2}{r}$$

---

$$\text{Total Energy} = E_{total} = E_{potential} + E_{kinetic}$$

For 2 orbiting bodies (e.g. Sun + Earth):  $E_{total} < 0$

For 2 unbound bodies (Earth + Mars rocket):  $E_{total} > 0$

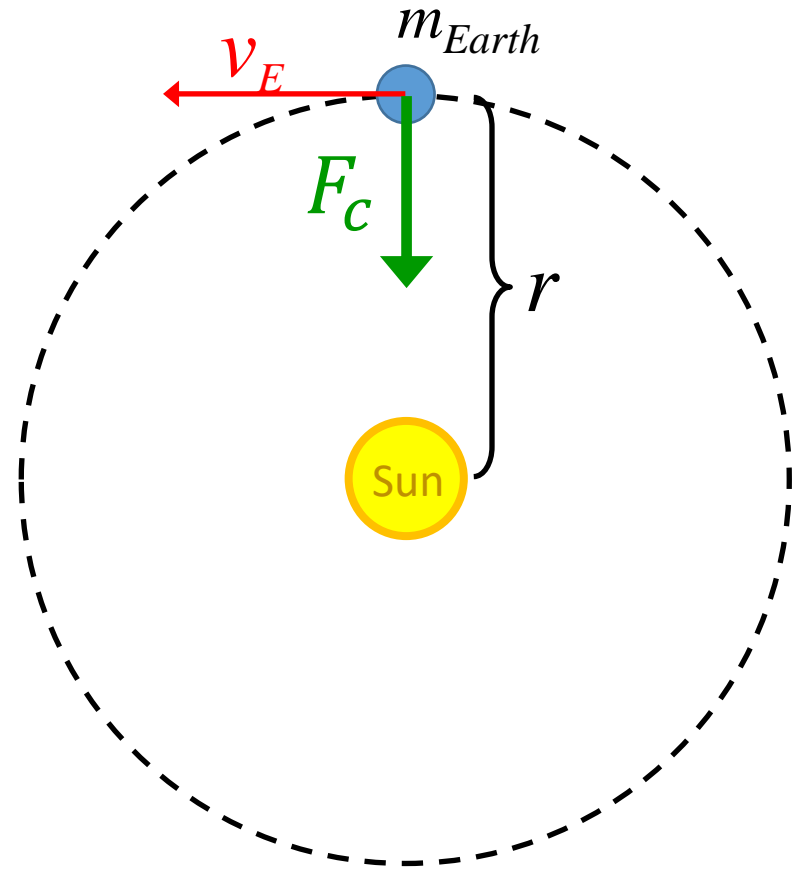
# Bound Orbital Energy

Centripetal force needed to keep Earth on a circular orbit:

$$F_c = \frac{m_{\text{Earth}} v_{\text{Earth}}^2}{r}$$

Force of gravity on Earth from Sun:

$$F_{\text{gravity}, S \rightarrow E} = G \frac{m_{\text{Earth}} M_{\text{Sun}}}{r^2}$$



# Bound Orbital Energy

Centripetal force needed to keep Earth on a circular orbit:

$$F_c = \frac{m_{\text{Earth}} v_{\text{Earth}}^2}{r}$$

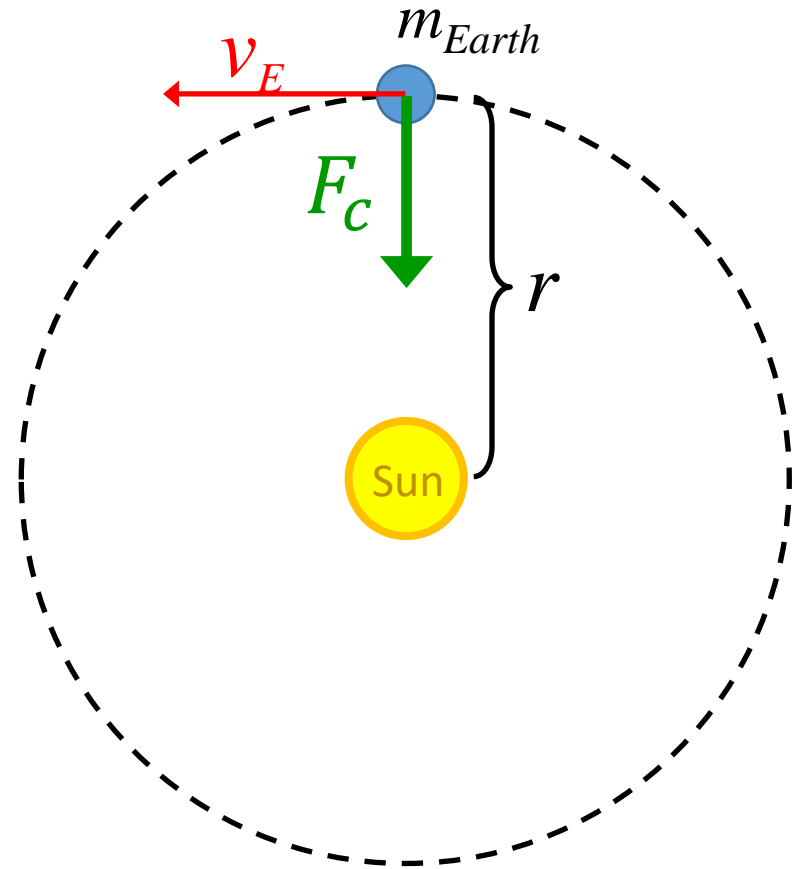
Force of gravity on Earth from Sun:

$$F_{\text{gravity}, S \rightarrow E} = G \frac{m_{\text{Earth}} M_{\text{Sun}}}{r^2}$$

The **centripetal force** that pulls on Earth to make it orbit the Sun **is gravity**:

$$F_c = F_{\text{gravity}, S \rightarrow E}$$

$$\Leftrightarrow \frac{m_{\text{Earth}} v_{\text{Earth}}^2}{r} = G \frac{m_{\text{Earth}} M_{\text{Sun}}}{r^2}$$





# Bound Orbital Energy

Centripetal force needed to keep Earth on a circular orbit:

$$F_c = \frac{m_{\text{Earth}} v_{\text{Earth}}^2}{r}$$

Force of gravity on Earth from Sun:

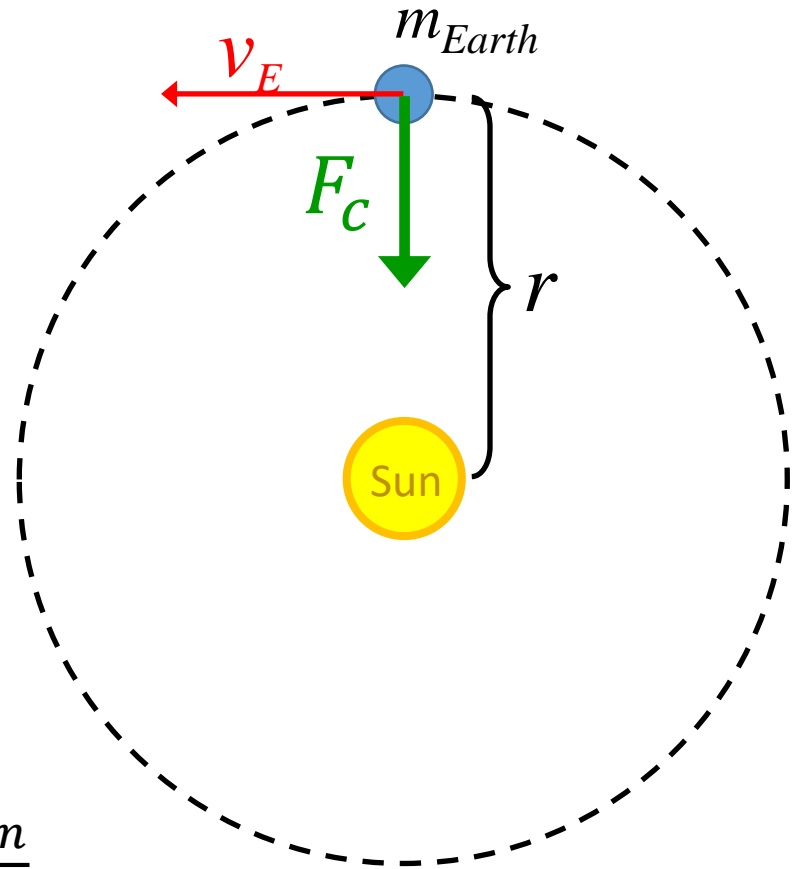
$$F_{\text{gravity}, S \rightarrow E} = G \frac{m_{\text{Earth}} M_{\text{Sun}}}{r^2}$$

The **centripetal force** that pulls on Earth to make it orbit the Sun **is gravity**:

$$F_c = F_{\text{gravity}, S \rightarrow E}$$

$$\Leftrightarrow \frac{m_{\text{Earth}} v_{\text{Earth}}^2}{r} = G \frac{m_{\text{Earth}} M_{\text{Sun}}}{r^2}$$

$$\Leftrightarrow \frac{1}{2} m_{\text{Earth}} v_{\text{Earth}}^2 = \frac{1}{2} G \frac{m_{\text{Earth}} M_{\text{Sun}}}{r}$$



# Bound Orbital Energy

Centripetal force needed to keep Earth on a circular orbit:

$$F_c = \frac{m_{\text{Earth}} v_{\text{Earth}}^2}{r}$$

Force of gravity on Earth from Sun:

$$F_{\text{gravity}, S \rightarrow E} = G \frac{m_{\text{Earth}} M_{\text{Sun}}}{r^2}$$

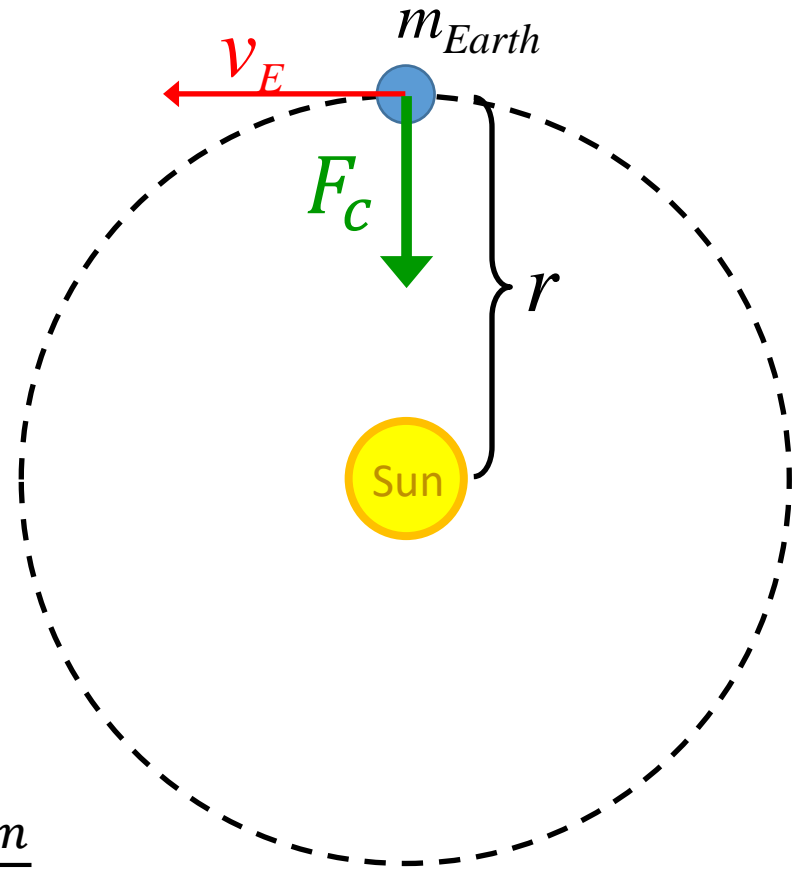
The **centripetal force** that pulls on Earth to make it orbit the Sun **is gravity**:

$$F_c = F_{\text{gravity}, S \rightarrow E}$$

$$\Leftrightarrow \frac{m_{\text{Earth}} v_{\text{Earth}}^2}{r} = G \frac{m_{\text{Earth}} M_{\text{Sun}}}{r^2}$$

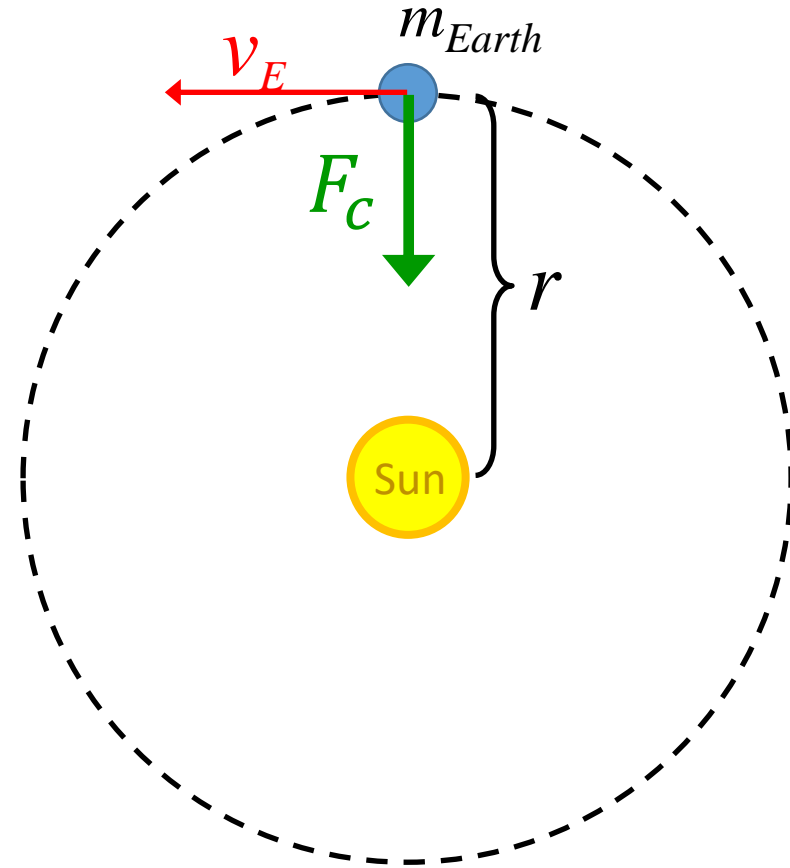
$$\Leftrightarrow \frac{1}{2} m_{\text{Earth}} v_{\text{Earth}}^2 = \frac{1}{2} G \frac{m_{\text{Earth}} M_{\text{Sun}}}{r}$$

$$\Leftrightarrow E_{\text{kinetic}} = \frac{1}{2} m_{\text{Earth}} v_{\text{Earth}}^2 = \frac{1}{2} G \frac{m_{\text{Earth}} M_{\text{Sun}}}{r}$$



# Bound Orbital Energy

$$E_{kinetic} = \frac{1}{2} G \frac{m_{Earth} M_{Sun}}{r}$$

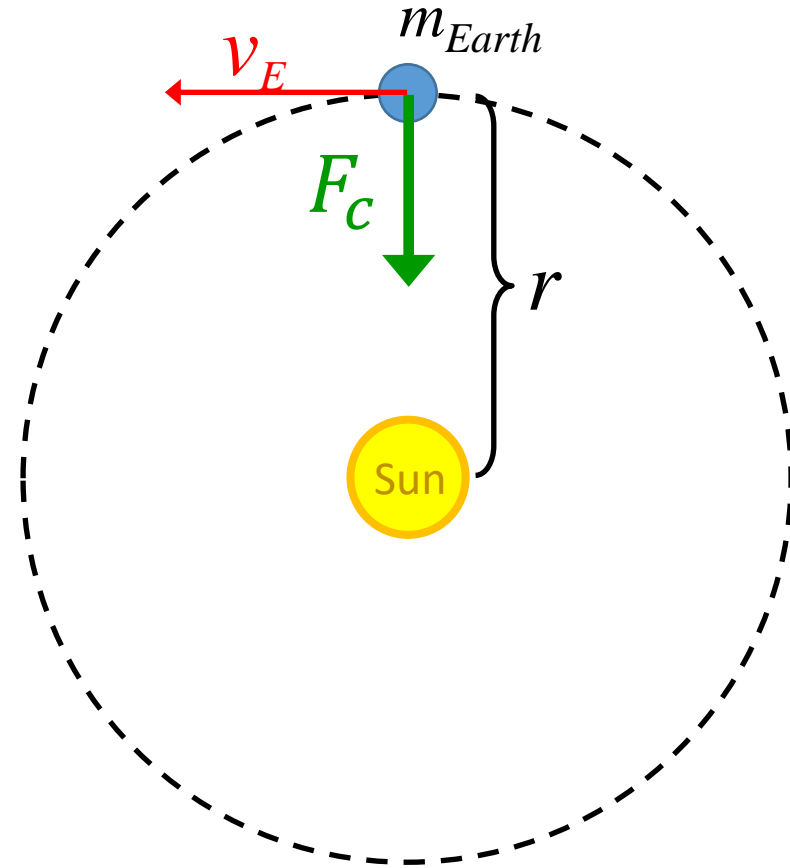


# Bound Orbital Energy

$$E_{kinetic} = \frac{1}{2} G \frac{m_{Earth} M_{Sun}}{r}$$

Total Energy:

$$E_{Total} = E_{kinetic} + E_{potential}$$

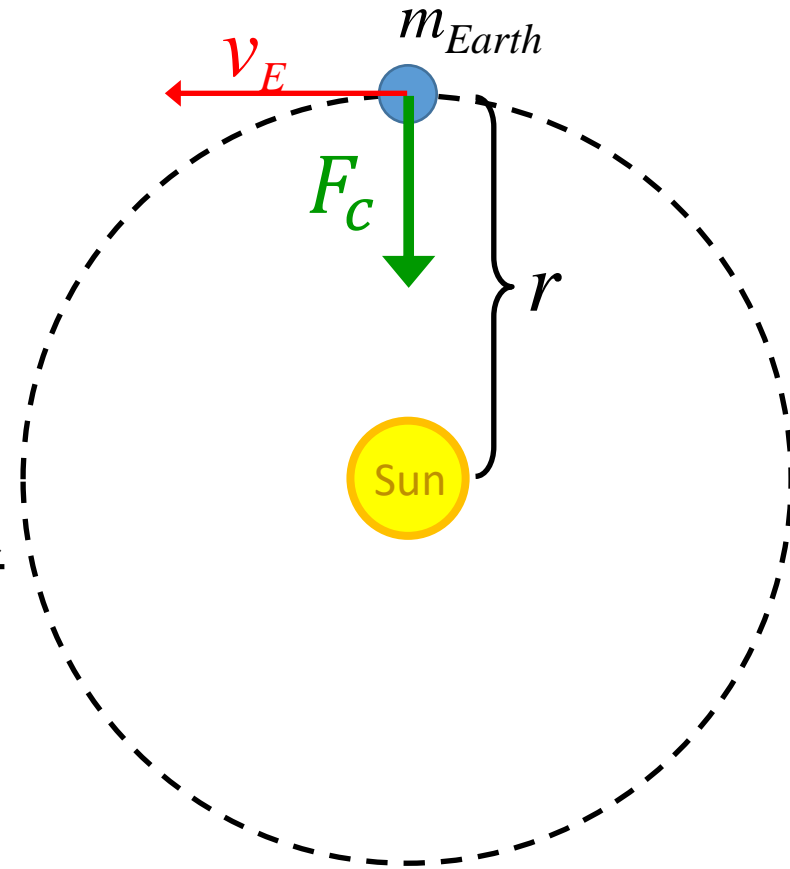


# Bound Orbital Energy

$$E_{kinetic} = \frac{1}{2} G \frac{m_{Earth} M_{Sun}}{r}$$

Total Energy:

$$\begin{aligned} E_{Total} &= E_{kinetic} + E_{potential} \\ &= \frac{1}{2} G \frac{m_{Earth} M_{Sun}}{r} - G \frac{m_{Earth} M_{Sun}}{r} \end{aligned}$$



# Bound Orbital Energy

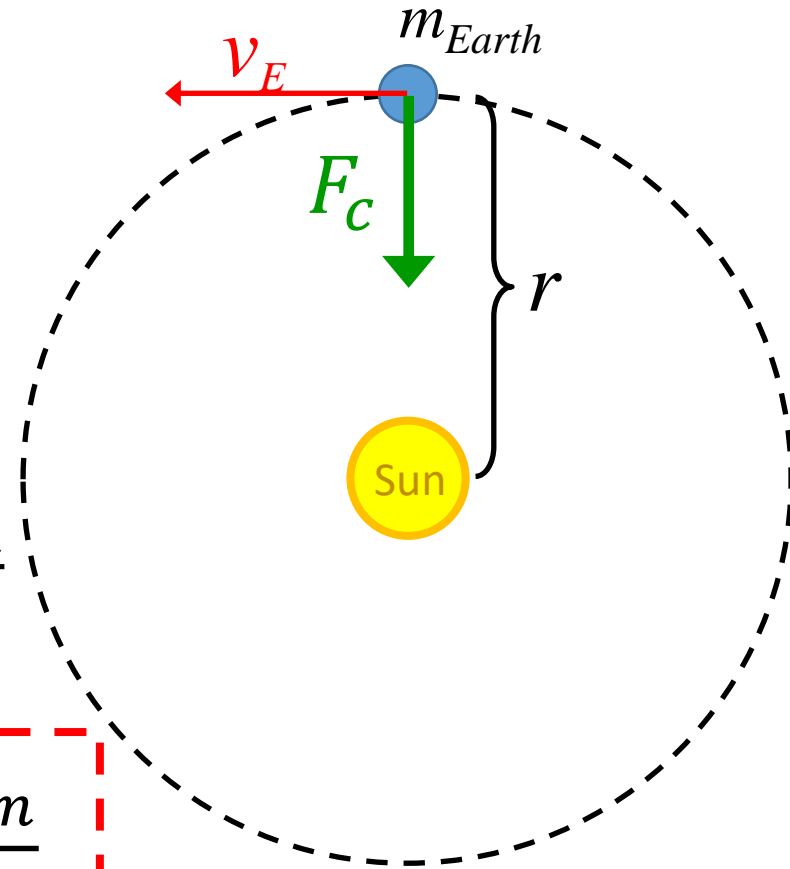
$$E_{kinetic} = \frac{1}{2} G \frac{m_{Earth} M_{Sun}}{r}$$

Total Energy:

$$E_{Total} = E_{kinetic} + E_{potential}$$

$$= \frac{1}{2} G \frac{m_{Earth} M_{Sun}}{r} - G \frac{m_{Earth} M_{Sun}}{r}$$

$$\Leftrightarrow E_{Total} = -\frac{1}{2} G \frac{m_{Earth} M_{Sun}}{r}$$



# Bound Orbital Energy

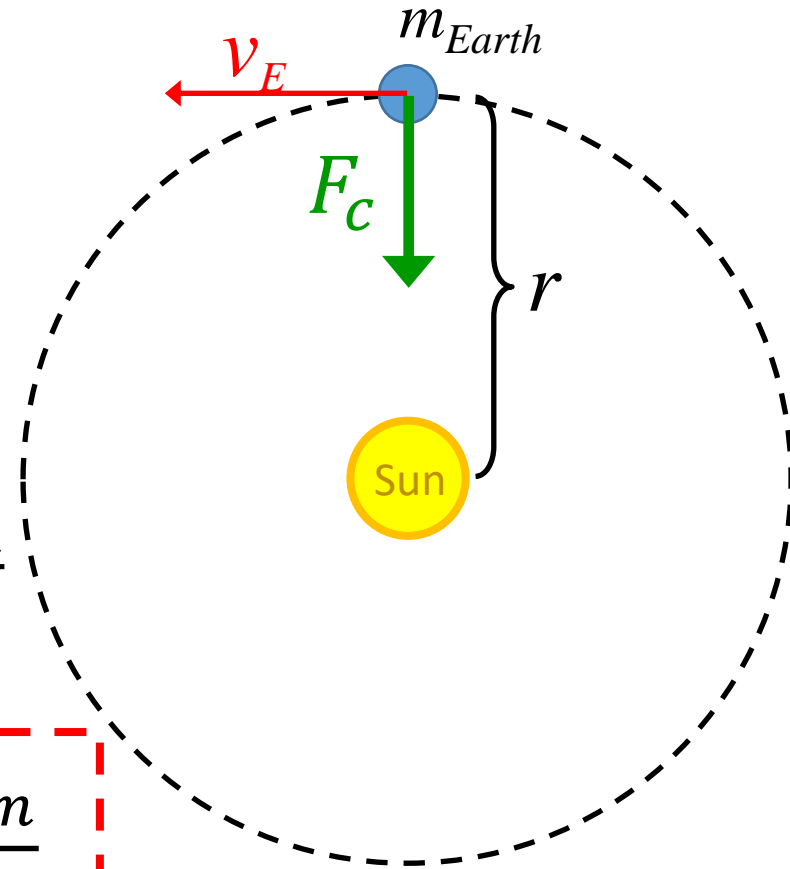
$$E_{kinetic} = \frac{1}{2} G \frac{m_{Earth} M_{Sun}}{r}$$

Total Energy:

$$E_{Total} = E_{kinetic} + E_{potential}$$

$$= \frac{1}{2} G \frac{m_{Earth} M_{Sun}}{r} - G \frac{m_{Earth} M_{Sun}}{r}$$

$$\Leftrightarrow E_{Total} = -\frac{1}{2} G \frac{m_{Earth} M_{Sun}}{r}$$



The bound orbital energy is negative:  $E_{Total} < 0$

*Example:* When a rocket wants to orbit another planet it has to slow (lower its energy) in order to go into orbit.